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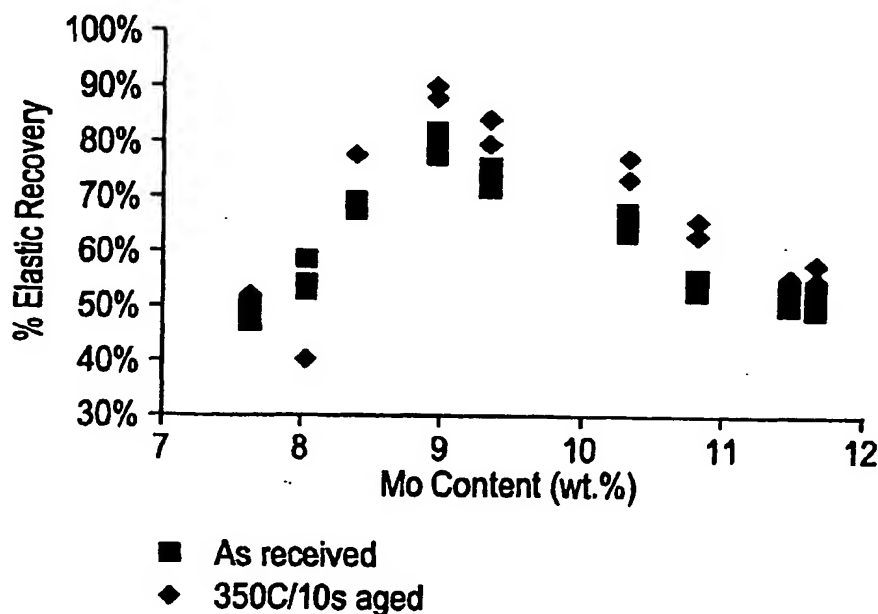
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(54) Title: B TITANIUM COMPOSITIONS AND METHODS OF MANUFACTURE THEREOF



(57) Abstract: A composition comprises about 8 to about 10 wt% molybdenum, about 2.8 to about 6 wt% aluminum, up to about 2 wt% vanadium, up to about 4 wt% niobium, with the balance being titanium, wherein the weight percents are based on the total weight of the alloy composition. A method for making an article comprises cold working a shape from a composition comprising about 8 to about 10 wt% molybdenum, about 2.8 to about 6 wt% aluminum, up to about 2 wt% vanadium, up to about 4 wt% niobium, with the balance being titanium, wherein the weight percents are based on the total weight of the alloy composition; solution heat treating the shape; and cooling the shape.

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## $\beta$ TITANIUM COMPOSITIONS AND METHODS OF MANUFACTURE THEREOF

### BACKGROUND

This disclosure relates to superelastic  $\beta$  titanium alloys, methods for manufacturing these alloys and articles derived therefrom.

Alloys that undergo a martensitic transformation may exhibit a “shape  
5 memory effect”. As a result of this transformation, the high temperature phase known  
as “austenite” changes its crystalline structure through a diffusion-less shear process  
adopting a less symmetrical structure called ‘martensite’. This process may be  
reversible as in shape memory alloys and therefore upon heating, the reverse  
transformation occurs. The starting temperature of the cooling or martensitic  
10 transformation is generally referred to as the  $M_s$  temperature and the finishing  
temperature is referred to as the  $M_f$  temperature. The starting and finishing  
temperatures of the reverse or austenitic transformation are referred to as  $A_s$  and  $A_f$   
respectively.

At temperatures below the  $A_f$ , alloys undergoing a reversible martensitic phase  
15 transformation may be deformed in their high temperature austenitic phase through a  
stress-induced martensitic transformation as well as in their low temperature  
martensitic phase. These alloys generally recover their original shapes upon heating  
above the  $A_f$  temperature and are therefore called “shape memory alloys”. At  
temperatures above the  $A_f$ , the stress-induced martensite is not stable and will revert  
20 back to austenite upon the release of deformation. The strain recovery associated with  
the reversion of stress-induced martensite back to austenite is generally referred to as  
“pseudoelasticity” or “superelasticity” as defined in ASTM F2005, Standard  
Terminology for Nickel-Titanium Shape Memory Alloys. The two terms are used  
interchangeably to describe the ability of shape memory alloys to elastically recover  
25 large deformations without a significant amount of plasticity due to the mechanically  
induced crystalline phase change.

Nitinol is a shape memory alloy comprising a near-stoichiometric amount of  
nickel and titanium. When deforming pseudoelastic nitinol, the formation of stress-

induced-martensite allows the strain of the alloy to increase at a relatively constant stress. Upon unloading, the reversion of the martensite back to austenite occurs at a constant, but different, stress. A typical stress-strain curve of pseudoelastic nitinol therefore exhibits both loading and unloading stress plateaus. However, since the stresses are different, these plateaus are not identical, which is indicative of the development of mechanical hysteresis in the nitinol. Deformations of about 8 to about 10% can thus be recovered in the pseudoelastic nitinol. Cold worked Nitinol also exhibits extended linear elasticity. Nitinol compositions, which display linear elasticity do not display any plateau but can recover a strain of up to 3.5%. This behavior is generally termed "Linear Superelasticity" to differentiate from transformation induced "Pseudoelasticity" or "Superelasticity". These properties generally make nitinol a widely used material in a number of applications, such as medical stents, guide wires, surgical devices, orthodontic appliances, cellular phone antenna wires as well as frames and other components for eye wear. However, nitinol is difficult to fabricate by forming and/or welding, which makes the manufacturing of articles from it expensive and time-consuming. Additionally, users of nickel containing products are sometimes allergic to nickel.

#### SUMMARY

In one embodiment, a composition comprises about 8 to about 10 wt% molybdenum, about 2.8 to about 6 wt% aluminum, up to about 2 wt% vanadium, up to about 4 wt% niobium, with the balance being titanium, wherein the weight percents are based on the total weight of the alloy composition.

In another embodiment, a composition comprises about 8.9 wt% molybdenum, about 3.03 wt% aluminum, about 1.95 wt% vanadium, about 3.86 wt% niobium, with the balance being titanium.

In yet another embodiment, a composition comprises about 9.34 wt% molybdenum, about 3.01 wt% aluminum, about 1.95 wt% vanadium, about 3.79 wt% niobium, with the balance being titanium.

In yet another embodiment, a method for making an article comprises cold working a shape from a composition comprising about 8 to about 10 wt% molybdenum, about 2.8 to about 6 wt% aluminum, up to about 2 wt% vanadium, up

to about 4 wt% niobium, with the balance being titanium, wherein the weight percents are based on the total weight of the alloy composition; forming the shape; solution heat treating the shape; and cooling the shape.

In yet another embodiment, a method comprises cold working a wire having a composition comprising about 8 to about 10 wt% molybdenum, about 2.8 to about 6 wt% aluminum, up to about 2 wt% vanadium, up to about 4 wt% niobium, with the balance being titanium, wherein the weight percents are based on the total weight of the alloy composition; cold working the shape; and heat treating the shape.

## 10 BRIEF DESCRIPTION OF THE DRAWINGS

Figure 1 is a graphical representation showing the effect of molybdenum content on elastic recovery;

Figure 2 is a graphical representation of the effect of aging at 350°C on the elastic recovery of Sample 4 from Table 1;

15 Figure 3 is a graphical representation of the effect of aging at 350°C on the elastic recovery of Sample 5 from Table 1;

Figure 4 is a graphical representation showing the effect of aging at 350°C on the elastic recovery of Sample 6 from Table 1;

20 Figure 5 is a graphic representation showing the effect of aging at about 250 to about 550°C for 10 seconds on the elastic recovery of Sample 4 from Table 1;

Figure 6 is a graphic representation showing the effect of aging at about 250 to about 550°C for 10 seconds on the elastic recovery of Sample 5 from Table 1;

Figure 7 is a graphical representation showing the effect of cumulative cold drawing reduction on the UTS of Sample 11 from Table 2;

5 Figure 8 is a graphical representation showing the effect of cumulative cold drawing reduction on the Young's Modulus of Sample 11 from Table 2;

Figure 9 is a graphical representation showing the effect of tensile stress-strain curve for a wire having the composition of Sample 11 from Table 2 with 19.4% drawing reduction, tested to 2% strain;

30 Figure 10 is a graphical representation showing the effect of tensile stress-strain curve for a wire having the composition of Sample 11 from Table 2 with 19.4% drawing reduction, tested to 4% strain;

Figure 11 is an optical micrograph showing the microstructure of a cold drawn wire having the composition of Sample 10 from Table 2 with a 14% reduction;

Figure 12 is an optical micrograph showing partially recrystallized microstructure of a cold-drawn wire having the composition of Sample 10 from Table 2 having a 14% reduction after heat- treating at 816°C for 30 minutes;

Figure 13 is an optical micrograph showing fully recrystallized microstructure of a cold-drawn wire having the composition of Sample 10 from Table 2 having a 14% reduction after heat- treating at 871°C for 30 minutes;

Figure 14 is an optical micrograph showing the microstructure of a betatized Sample 10 from Table 2 after aging at 816°C for 30 minutes;

Figure 15 is an optical micrograph showing the microstructure of a betatized Sample 10 from Table 2 after aging at 788°C for 30 minutes;

Figure 16 is a graphical representation showing the UTS of betatized Sample 10 from Table 2 after aging at 500-900°C for 30 minutes;

Figure 17 is a graphical representation showing the ductility of betatized Sample 10 from Table 2 after aging at 500-900°C for 30 minutes;

Figure 18 is a graphical representation showing a tensile stress-strain curve tested to 4% tensile strain of a wire having the composition of Sample 11 from Table 2 after strand annealing at 871°C; and

Figure 19 is an optical micrograph showing the microstructure of a wire having the composition of Sample 11 from Table 2 after strand annealing at 871°C.

## DETAILED DESCRIPTION OF PREFERRED EMBODIMENTS

Disclosed herein is a  $\beta$  titanium alloy composition having pseudoelastic properties and linear-superelastic properties that can be used for medical, dental, sporting good and eyewear frame applications. In one embodiment, the  $\beta$  titanium alloy composition has linear elastic properties after solution treatment. In another embodiment, the  $\beta$  titanium alloy composition has pseudoelastic properties that are improved with heat treatment. In yet another embodiment, the  $\beta$  titanium alloy composition displays linear-superelastic properties after being cold worked. The composition advantageously can be welded to other metals and alloys. The articles manufactured from the  $\beta$  titanium alloy can also be deformed into various shapes at ambient temperature and generally retain the high spring back characteristics associated with superelasticity.

Pure titanium has an isomorphous transformation temperature at 882°C. The body centered cubic (bcc) structure, which is called  $\beta$ -titanium, is stable above the isomorphous transformation temperature, and the hexagonal close packed (hcp) structure, which is called  $\alpha$  titanium is generally stable below this temperature. When titanium is alloyed with elements such as vanadium, molybdenum, and/or niobium, the resulting alloys have an increased  $\beta$  phase stability at temperatures less than or equal to about 882°C ( $\beta$  transus temperature). On the other hand, when alloyed with elements such as aluminum or oxygen, the temperature range of the stable  $\alpha$  phase is increased above the isomorphous transformation temperature. Elements which have the effect of increasing the  $\beta$  phase temperature range are called the  $\beta$  stabilizers, while those capable of extending the  $\alpha$  phase temperature range are called the  $\alpha$  stabilizers.

Titanium alloys having a high enough concentration of  $\beta$  stabilizers, generally are sufficiently stable to have a meta-stable  $\beta$  phase structure at room temperature. The alloys showing such a property are called  $\beta$  titanium alloys. Martensite transformations are generally present in  $\beta$  titanium alloys. The martensitic transformation temperature in  $\beta$  titanium alloys generally decreases with an increasing amount of  $\beta$  stabilizer in the alloy, while increasing the amount of  $\alpha$  stabilizer generally raises the martensitic transformation temperature. Therefore, depending on the extent of stabilization,  $\beta$  titanium alloys may exhibit a martensitic transformation

when cooled rapidly from temperatures greater than those at which the  $\beta$  phase is the single phase at equilibrium.

The  $\beta$  titanium alloy generally comprises an amount of about 8 to about 10 wt% of molybdenum, about 2.8 to about 6 wt% aluminum, up to about 2 wt% vanadium, up to about 4 wt% niobium, with the balance being titanium. All weight percents are based on the total weight of the alloy. Within the aforementioned range for molybdenum, it is generally desirable to have an amount of greater than or equal to about 8.5, preferably greater than or equal to about 9.0, and more preferably greater than or equal to about 9.2 wt% molybdenum. Also desirable within this range is an amount of less than or equal to about 9.75, preferably less than or equal to about 9.65, and more preferably less than or equal to about 9.5 wt% molybdenum, based on the total weight of the alloy.

Within the aforementioned range for aluminum, it is generally desirable to have an amount of greater than or equal to about 2.85, preferably greater than or equal to about 2.9, and more preferably greater than or equal to about 2.93 wt% aluminum. Also desirable within this range is an amount of less than or equal to about 5.0, preferably less than or equal to about 4.5, and more preferably less than or equal to about 4.0 wt% aluminum, based on the total weight of the alloy.

Within the aforementioned range for niobium, it is generally desirable to have an amount of greater than or equal to about 2, preferably greater than or equal to about 3, and more preferably greater than or equal to about 3.5 wt% niobium, based on the total weight of the alloy.

In one exemplary embodiment, it is generally desirable for the  $\beta$  titanium alloy to comprise 8.9 wt% molybdenum, 3.03 wt% aluminum, 1.95 wt% vanadium, 3.86 wt% niobium, with the balance being titanium.

In another exemplary embodiment, it is generally desirable for the  $\beta$  titanium alloy to comprise 9.34 wt% molybdenum, 3.01 wt% aluminum, 1.95 wt% vanadium, 3.79 wt% niobium, with the balance being titanium.

In one embodiment, the  $\beta$  titanium alloy may be solution treated and/or thermally aged. In solution treating the  $\beta$  titanium alloy, the alloy is subjected to a temperature greater than or equal to about 850°C, the  $\beta$  transus temperature for the alloy. The solution treatment of the alloy is normally carried out in either vacuum or

inert gas environment at a temperature of about 850 to about 1000°C, preferably about 850 to about 900°C, for about 1 minute or longer in duration depending on the mass of the part. The heating is followed by a rapid cooling at a rate greater than or equal to about 5°C/second, preferably greater than or equal to about 25°C/second, and more preferably greater than or equal to about 50°C/second, by using an inert gas quench or air cooling to retain a fully recrystallized single phase  $\beta$  grain structure. In some instances, it is preferred that the quenched alloy is subsequently subjected to an ageing treatment at about 350 to about 550°C for about 10 seconds to about 30 minutes to adjust the amount of a fine precipitate of the  $\omega$  phase.

10 In another embodiment, the  $\beta$  titanium alloy may be solution treated at a temperature below the  $\beta$  transus temperature of about 750 to about 850°C, preferably about 800 to about 850°C, for about 1 to about 30 minutes to induce a small amount of  $\alpha$  precipitates in the recrystallized  $\beta$  matrix. The amount of the  $\alpha$  precipitates is preferably less than or equal to about 15 volume percent and more preferably less than  
15 or equal to about 10 volume percent, based on the total volume of the composition. This improves the tensile strength to an amount of greater than or equal to about 140,000 pounds per square inch (9,846 kilogram/square centimeter).

The  $\beta$  titanium alloy in the solution treated condition may exhibit pseudoelasticity. The solution treated  $\beta$  titanium alloy generally exhibits a  
20 pseudoelastic recovery of greater than or equal to about 75% of the initial strain when elastically deformed to a 2% initial strain, and greater than or equal to about 50% of the initial strain when elastically deformed to a 4% initial strain. The initial strain is the ratio of the change in length to the original length of the alloy composition.

The  $\beta$  titanium alloy in the solution treated condition may exhibit linear  
25 elasticity. The solution treated  $\beta$  titanium alloy generally exhibits a linear elastic recovery of greater than or equal to about 75% of the initial strain when elastically deformed to a 2% initial strain, and greater than or equal to about 50% of the initial strain when elastically deformed to a 4% initial strain. The initial strain is the ratio of the change in length to the original length of the alloy composition.

30 In another embodiment, the  $\beta$  titanium alloy may be cold worked by processes such as cold rolling, drawing, swaging, pressing, and the like, at ambient



temperatures. The  $\beta$  titanium alloy may preferably be cold worked to an amount of about 5 to about 85% as measured by the reduction in cross-sectional area based upon the original cross sectional area. Within this range it is desirable to have a cross sectional area reduction of greater than or equal to about 10, preferably greater than or equal to about 15% of the initial cross sectional area. Also desirable within this range is a reduction in cross sectional area of less than or equal to about 50, more preferably less than or equal to about 30% based on the initial cross-sectional area. The  $\beta$  titanium alloy in the cold worked state (also referred to as the work hardened state) exhibits linear superelasticity where greater than or equal to about 75% of the initial strain is elastically recoverable after deforming to a 2% initial strain, and greater than or equal to about 50% of the initial strain is elastically recoverable after deforming to a 4% initial strain. In one exemplary embodiment related to cold working, the elastic modulus of the  $\beta$  titanium alloy is reduced through cold working by an amount of greater than or equal to about 10, preferably greater than or equal to about 20 and more preferably greater than or equal to about 25% based upon the elastic modulus, after the alloy is heat treated.

It is generally desirable to use shape memory alloys having pseudo-elastic properties, and which are formable into complex shapes and geometries without the creation of cracks or fractures. In one embodiment, the  $\beta$  titanium alloy having linear elastic, linearly superelastic, pseudoelastic or superelastic properties may be used in the manufacturing of various articles of commerce. Suitable examples of such articles are eyewear frames, face inserts or heads for golf clubs, medical devices such as orthopedic prostheses, spinal correction devices, fixation devices for fracture management, vascular and non-vascular stents, minimally invasive surgical instruments, filters, baskets, forceps, graspers, orthodontic appliances such as dental implants, arch wires, drills and files, and a catheter introducer (guide wire).

The superelastic  $\beta$  titanium alloy generally provides an adequate spring-back for eyewear applications. It is generally desired to use superelastic  $\beta$  titanium alloy having a minimum recovery of about 50% of the initial strain, when the alloy is deformed to an outer fiber initial strain of about 4% in a bend test. It is preferable to have a minimum recovery of greater than or equal to about 75% of the initial strain when the alloy composition is deformed to about 4% of the outer fiber initial length in

a bend test. It is also generally desirable for the superelastic  $\beta$  titanium alloy to have a minimum recoverable strain of about 50% of the initial strain, when the alloy composition is strained to about 4% initial tensile strain. It is preferable to have a minimum recovery of greater than or equal to about 75% of the initial tensile strain, when the alloy is strained to about 4% initial strain in a tensile test. The strain recovery is measured as a function of the initial bending strain and the initial bending strain is expressed as a percentage of the ratio of the change in length to the original length.

The following examples, which are meant to be exemplary, not limiting, illustrate some of the various embodiments of the  $\beta$  titanium alloy compositions described herein.

#### EXAMPLES

##### Example 1

All of the sample alloys discussed below were prepared by a double vacuum arc melting technique. The ingots were hot rolled and flattened to sheets having a thickness of 1.5 millimeter (mm). The sheets were then heat treated at 870°C for 30 minutes in air and air cooled to ambient temperature. Oxides on the sheets were removed by double-disc grinding and lapping to a thickness of 1.3 mm. Heat aging experiments were conducted at 350°C using a nitride/nitrate salt bath.

Permanent deformation and pseudo-elastic recovery strains were determined using bend tests. Specimens having dimensions 0.51 mm x 1.27 mm x 51 mm were cut from the sheets. The specimens were bent against a rod of approximately 12.2 mm in diameter to form a "U" shape to yield an outer fiber or outer surface strain close to 4%. The angles between the straight portions were measured afterwards and the strain recovery calculated by using the formula:

$$e(\text{rec}) = e(180 - a)/180;$$

where "a" is the unrecovered angle and "e" is the outer-fiber bending strain.

Tensile strain recovery was measured by tensile elongation to a strain of 4% followed by unloading to zero stress. Tensile specimens with a cross sectional dimension of 0.90 mm x 2.0 mm were used and the strain was monitored using an extensometer. An environmental chamber with electrical heating and CO<sub>2</sub> cooling capabilities provided a testing capability from -30°C to 180°C.

Nine  $\beta$  titanium alloys having the compositions listed in Table 1 were examined. The percentage of the elastic recovery strain with respect to the total bend strain was measured for comparison.

Table 1

Sample #	Titanium	Molybdenum	Niobium	Vanadium	Aluminum
1	Balance	7.63	3.98	2.05	3.10
2	Balance	8.03	3.89	2.03	3.09
3	Balance	8.40	3.83	1.94	3.03
4	Balance	8.97	3.86	1.95	3.03
5	Balance	9.34	3.79	1.95	3.01
6	Balance	10.35	3.83	1.99	3.02
7	Balance	10.83	3.88	2.01	3.02
8	Balance	11.48	4.00	2.04	3.15
9	Balance	11.68	3.89	1.98	3.07

5

In the Table 1 above Sample 1 and Samples 6 – 9 are comparative examples. The results of elastic recovery after bending to approximately 4% outer fiber strain is graphically demonstrated in Figure 1. The figure shows a maximum elastic strain recovery at about 9 wt% molybdenum, where the alloy after solution heat treatment and subsequent air cooling, exhibits an elastic recovery strain of approximately 80% of the applied 4% deformation strain. Increasing or decreasing the molybdenum content from 9 wt% generally results in decreasing elastic recovery. It may also be seen that an aging treatment at 350°C for a short duration of 10 seconds results in an improved elastic recovery, for titanium alloys having a molybdenum content between 8.4 and 11 wt%. The optimal elastic strain recovery after heat aging at 350°C for 10 seconds for the alloy having about 9 wt% molybdenum is approximately 90% of the applied 4% bend strain. Alloys with a molybdenum content less than 8.4 wt% exhibit

10

15

a different aging characteristic. Aging at 350°C may degrade elastic strain recovery as for alloy 2 having about 8.03 wt% molybdenum, or has no significant effect as for alloy 1 having about 7.63 wt% molybdenum.

5 The percent of the elastic recovery to the total deformation during thermal aging at 350°C for Samples 4, 5 and 6 respectively, are plotted in the Figures 2, 3 and 4 respectively. From the Figures 2, 3 and 4 it may be seen that the elastic recoveries of all three alloys reach a maximum after aging for about 10 to about 60 seconds. Aging beyond 15 minutes (900 seconds) degrades the elastic recovery.

10 The percents of the elastic recovery to the total deformation during thermal aging at about 250 to about 550°C for 10 seconds for Samples 4 and 5 respectively are plotted in the Figures 5 and 6, respectively. An optimal for Sample 4 appears at 350°C, which improves the elastic recovery to a percentage close to 90% while aging at temperatures equal to or higher than 400°C degrade elastic recovery to about 40%. For Sample 5, aging in this temperature range generally improves the elastic recovery.  
15 The maximum improvement occurs at about 450°C where the elastic recovery is improved to 90%.

The alloys shown in Table I also exhibit linear superelasticity after cold working with a reduction of greater than or equal to about 30% in the cross-sectional area. For example, a wire fabricated from an ingot having a composition of 11.06  
20 wt% molybdenum, 3.80 wt% niobium, 1.97 wt% vanadium, 3.07 wt% aluminum with the remainder being titanium exhibited an elastic recovery strain of 3.5% after bending to a total deformation of 4% outer fiber strain, when the reduction in the cross sectional area after cold working was 84%.

## 25 Example 2

In this example, the  $\beta$  titanium alloys were manufactured by double vacuum arc melting. Chemistries of the alloys were analyzed using inductively coupled plasma optical emission spectrometry (ICP-OE). The results are tabulated in Table 2. The ingot was hot-forged, hot-rolled and finally cold-drawn to wire of various  
30 diameters in the range of about 0.4 to about 5 mm. Inter-pass annealing between cold reductions was carried out at 870°C in a vacuum furnace for wires having a diameter

of larger than 2.5 mm or by strand annealing under inert atmosphere for the smaller diameters. Tensile properties were determined using an Instron model 5565 material testing machine equipped with an extensometer of 12.5 mm gage length. Microstructures were studied by optical metallography using a Nikon Epiphot inverted metallurgical microscope.

Table 2

Sample #	Titanium	Molybdenum	Niobium	Vanadium	Aluminum	MoEq
10	Balance	11.06	3.80	1.97	3.07	10.37
11	Balance	9.59	3.98	1.99	3.13	8.91

The strand-annealed wires generally have a higher ultimate tensile strength (UTS) around 1055 mega Pascals (MPa) than vacuum annealed wires and sheets, the typical UTS of which is about 830 MPa. Figure 7 plots the UTS of wires drawn from an annealed 1.0 mm diameter Sample 11 wire stock as a function of reduction in cross-section area. After a 49% reduction, the UTS was elevated from 1055 MPa to only 1172 MPa indicating a fairly weak strain hardening effect. Young's Modulus was determined by tensile testing the wire to 1% strain and measuring the linear slope of the stress-strain curve. As shown in Figure 8, cold-drawn wires generally have a lower modulus than does annealed wire. The modulus, of approximately 65.9 gigapascals (GPa) for the annealed wire, decreases with increasing accumulative amount of reduction and stabilizes at approximately 50 GPa after cold drawing with a cumulative reduction greater than 20%.

Similar to alloys in Table 1, Samples 10 and 11 exhibit linear superelasticity after cold working. Loading and unloading stress-strain curves tested to 2% and 4% tensile strains of a cold drawn, 0.91 mm diameter wire of Sample 11 with a 19.4% reduction are plotted in Figures 9 and 10, respectively. As may be seen in Figure 7, after unloading, following a 2% tensile elongation, the wire recovers the majority of the deformation leaving only a small plastic deformation of 0.1% strain. When deformed to a 4% tensile elongation, the residual strain after unloading increases to 1.4%. The wire recovers a strain of 2.6%. The residual strain decreases with increasing drawing (cross-sectional area) reduction. However, when the reduction

exceeds 20%, specimens failed before reaching a 4% tensile elongation. As this data suggests, cold drawn  $\beta$  titanium alloy wires exhibit linear superelasticity and are capable of recovering large deformations greater than the typical elastic limit for conventional metallic alloys. The mechanical property of cold-drawn wire appears to be insensitive to chemical composition as the cold-drawn Sample 10 exhibits similar mechanical properties. All the loading/unloading tensile test results for Sample 10 are tabulated in Tables 3.

Table 3

Work Amount (%)	21	37	50	61	69
Tested to 2% tensile strain					
Elastic Strain (%)	1.9	1.8	1.8	1.9	2.0
Plastic Strain (%)	0.1	0.2	0.2	0.1	0.0
Tested to 3% tensile strain					
Elastic Strain (%)	2.5	2.6	2.6	2.7	2.7
Plastic Strain (%)	0.5	0.4	0.4	0.3	0.3
Tested to 4% tensile strain					
Elastic Strain (%)	---	2.8	2.9	3.1	3.2
Plastic Strain (%)	---	1.2	1.1	0.9	0.8

micrograph in Figure 11 reveals the cold-worked microstructure of the Sample 10 wire after a 14% cold working reduction in cross sectional area. The recrystallized microstructures of the wire after heat-treatments at 816°C and 871°C for 30 minutes are shown in Figures 12 and 13, respectively. It is apparent that the material was not fully betatized after the heat-treatment at 816°C as  $\alpha$  phase was present in the microstructure. As may be seen in Figure 11, a fully recrystallized  $\beta$  grain structure was obtained after the heat-treatment at 871°C for 30 minutes.

Sample 10 wires hot-rolled to 8.6 mm in diameter were further drawn down to 6.0 mm diameter. After being fully betatized at 871°C for 30 minutes the 6.0 mm diameter wires were again aged at temperatures of about 500 to about 850°C for 30 minutes. As can be seen in Figure 14, the  $\beta$  structure was preserved after aging at 816°C. When the aging temperature was lowered to 788°C, intragranular  $\alpha$ -phase

precipitates began to appear in the microstructure as may be seen in Figure 15. The amount of intragranular  $\alpha$ -phase precipitate increased with decreasing aging temperature.  $\alpha$ -phase precipitates eventually appeared along the grain boundary when aged at 649°C and below.

5           The ultimate tensile strength (UTS) and tensile ductility (% reduction in cross-section area) of betatized Sample 10 from Table 2 after aging at a temperature of about 500 to about 900°C for 30 minutes are plotted in Figures 16 and 17, respectively. Fully betatized specimens such as solution-treated specimens and those aged at 816°C and above, exhibited a low UTS of about 800 MPa and a good tensile  
10       ductility of about 25 to about 30% in reduction in cross-section area (RA). As the aging temperature decreased, there was a drastic increase in UTS with a significant reduction in tensile ductility, presumably due to an increasing amount of  $\alpha$ -precipitates. The peak of 1400 MPa in UTS coincides with the low in ductility (5% RA) and both appeared at approximately 500°C of aging temperature.

15           The Sample 11 composition in solution treated condition exhibits pseudoelasticity. Their mechanical properties are highly sensitive to solution heat treatment and subsequent aging at a temperature of about 350 to about 550°C. It was discovered that Sample 11 wires after strand annealing at 870°C exhibit well-defined pseudoelasticity. An example is presented in Figure 18, which shows a 4% tensile  
20       stress-strain curve of a strand-annealed, 0.4mm diameter Sample 11 wire. After deforming to a 4% elongation, the wire specimen was able to go through a pseudoelastic recovery recovering a 3.4% tensile strain and leaving a residual strain of only 0.6% after unloading.

          A transverse cross-sectional view of the wire microstructure is shown in a  
25       micrograph of Figure 19. Instead of the anticipated  $\beta$  structure, the microstructure consists of equiaxial  $\alpha$  precipitates in  $\beta$  matrix. It appears that the short duration of strand annealing did not allow the wire to fully recrystallize into the  $\beta$  grain structure. Without being limited by theory, it is believed that this may explain why strand-annealed wire generally has a higher UTS when compared to that of a fully betatized  
30       material.

As may be seen from the above experiments, the  $\beta$  titanium alloys can display an elastic strain recovery of 88.5%, when subjected to an initial bending strain of 4%. The strain recovery is measured as a function of the initial bending strain and the initial bending strain is expressed as a percentage of the ratio of the change in length to the original length. These alloys may be advantageously used in a number of commercial applications such as eyewear frames, face insert and heads for golf clubs, orthodontic arch wires, orthopedic prostheses and fracture fixation devices, spinal fusion and scoliosis correction instruments, stents, a catheter introducer (guide wire) and the like.

10 While the invention has been described with reference to exemplary embodiments, it will be understood by those skilled in the art that various changes may be made and equivalents may be substituted for elements thereof without departing from the scope of the invention. In addition, many modifications may be made to adapt a particular situation or material to the teachings of the invention  
15 without departing from the essential scope thereof. Therefore, it is intended that the invention not be limited to the particular embodiment disclosed as the best mode contemplated for carrying out this invention.

What is claimed is



1. A composition comprising  
about 8 to about 10 wt% molybdenum,  
about 2.8 to about 6 wt% aluminum,  
up to about 2 wt% vanadium,  
5 up to about 4 wt% niobium, with the balance being titanium, wherein the  
weight percents are based on the total weight of the alloy composition.
2. The composition of Claim 1, wherein the composition is cold worked.
3. The composition of Claim 2, wherein the composition, after cold working, has  
an elastic recovery of greater than or equal to about 75% of the applied change in  
length when the applied change in length is 2% of the original length.
4. The composition of Claim 2, wherein the composition, after cold working, has  
an elastic recovery of greater than or equal to about 50% of the applied change in  
length when the applied change in length is 4% of the original length.
5. The composition of Claim 2, wherein the composition, after cold working, has  
a reduction in the elastic modulus of greater than or equal to about 10% when  
compared with the elastic modulus of an equivalent heat treated composition.
6. The composition of Claim 2, wherein the composition has a  $\beta$  phase or an  $\alpha$   
phase and a  $\beta$  phase.
7. The composition of Claim 6, further comprising solution treating the  
composition.
8. An article manufactured from the composition of Claim 1.
9. A composition comprising about 8.9 wt% molybdenum, about 3.03 wt%  
aluminum, about 1.95 wt% vanadium, about 3.86 wt% niobium, with the balance  
being titanium.

10. The composition of Claim 9, wherein the composition after cold working has an elastic recovery of greater than or equal to about 75% of the applied change in length when the applied change in length is 2% of the original length.
11. A composition comprising about 9.34 wt% molybdenum, about 3.01 wt% aluminum, about 1.95 wt% vanadium, about 3.79 wt% niobium, with the balance being titanium.
12. The composition of Claim 11, wherein the composition after cold working has an elastic recovery of greater than or equal to about 50% of the applied change in length when the applied change in length is 4% of the original length.
13. A method for making an article comprising:  
cold working a shape from a composition comprising about 8 to about 10 wt% molybdenum, about 2.8 to about 6 wt% aluminum, up to about 2 wt% vanadium, up to about 4 wt% niobium, with the balance being titanium, wherein the weight percents  
5 are based on the total weight of the alloy composition;  
solution heat treating the shape; and  
cooling the shape.
14. The method of Claim 13, wherein the solution heat treating is conducted at a temperature below the isomorphic temperature for the composition or at a temperature above the isomorphic temperature for the composition.
15. The method of Claim 13, wherein the cooling is conducted in air.
16. The method of Claim 13, wherein the shape is further heat aged at a temperature of about 350 to about 550°C.
17. The method of Claim 13, wherein the heat ageing is conducted for a time period of 10 seconds to about 30 minutes.

18. A method comprising:  
cold working a wire having a composition comprising about 8 to about 10 wt% molybdenum, about 2.8 to about 6 wt% aluminum, up to about 2 wt% vanadium, up to about 4 wt% niobium, with the balance being titanium, wherein the weight  
5 percents are based on the total weight of the alloy composition;  
solution treating the wire; and  
heat treating the wire.
19. The method of Claim 18, wherein the cold working results in a reduction in cross-sectional area of about 5 to about 85%.
20. The method of Claim 18, wherein the heat treating is conducted at a temperature of about 500°C to about 900°C.
21. The method of Claim 18, wherein the wire is solution treated at a temperature of about 800 to about 1000°C.
22. The method of Claim 18, wherein the article has a  $\beta$  phase or an  $\alpha$  phase and a  $\beta$  phase.
23. An article manufactured by the method of Claim 13 or 18.

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FIG. 1

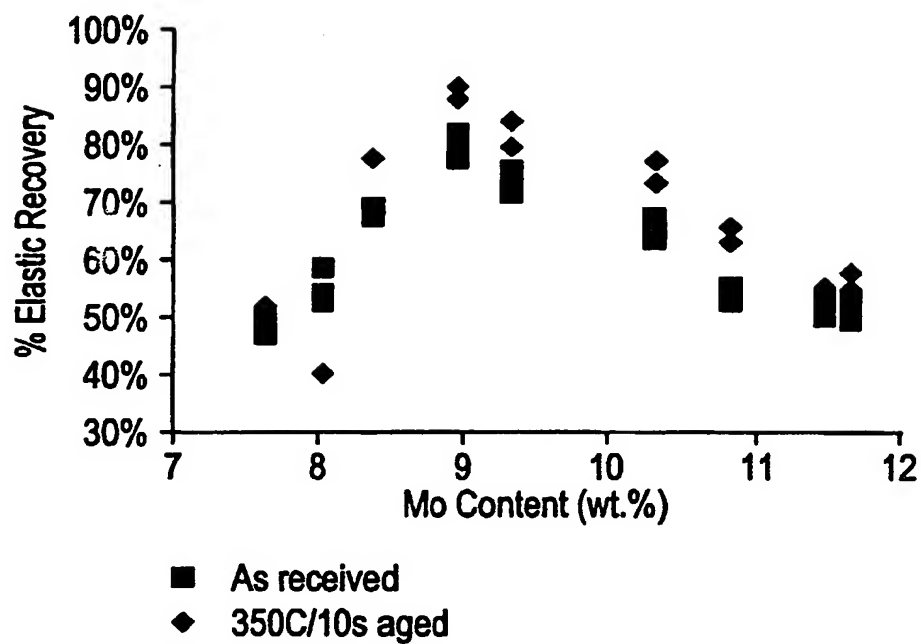
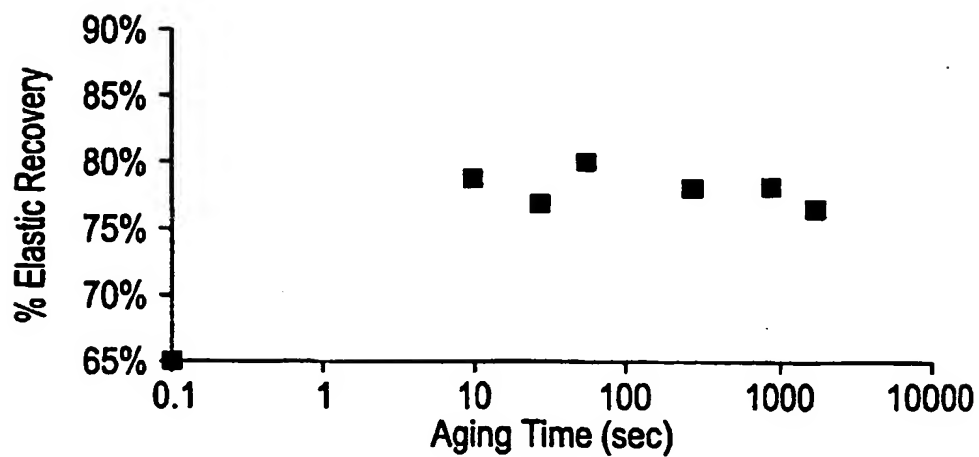


FIG. 2



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FIG. 3

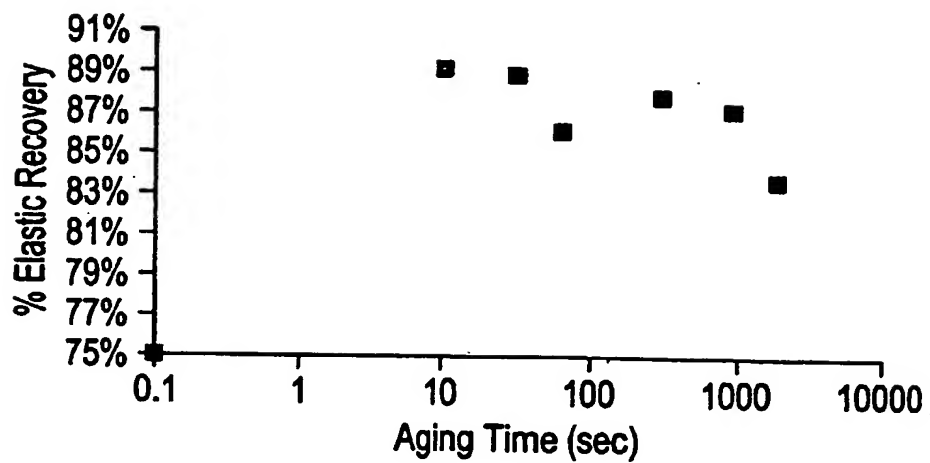
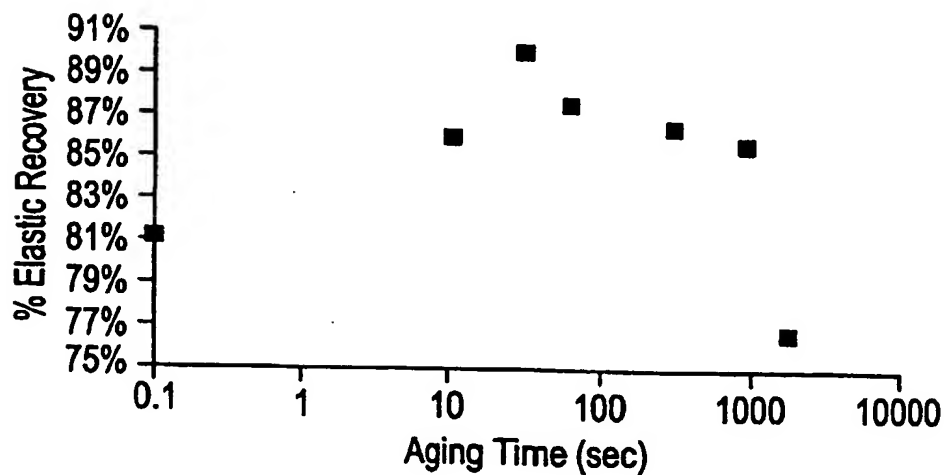


FIG. 4



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FIG. 5

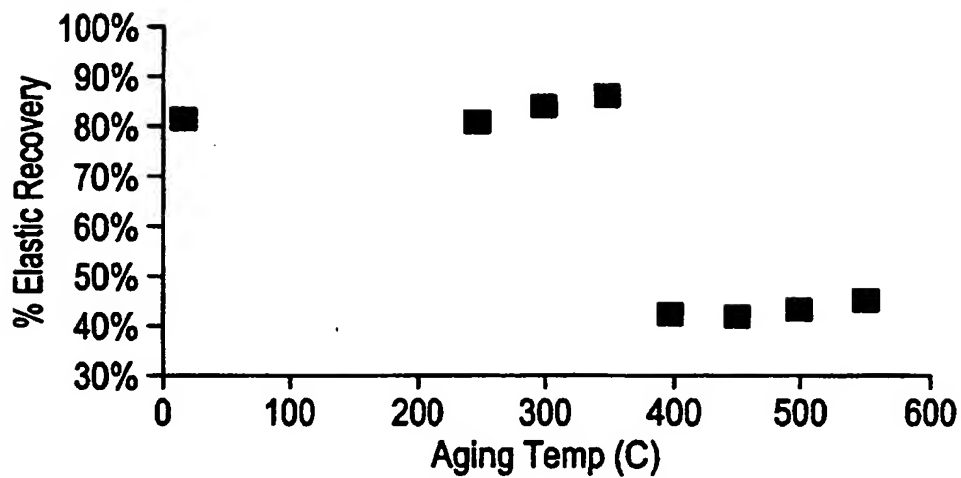
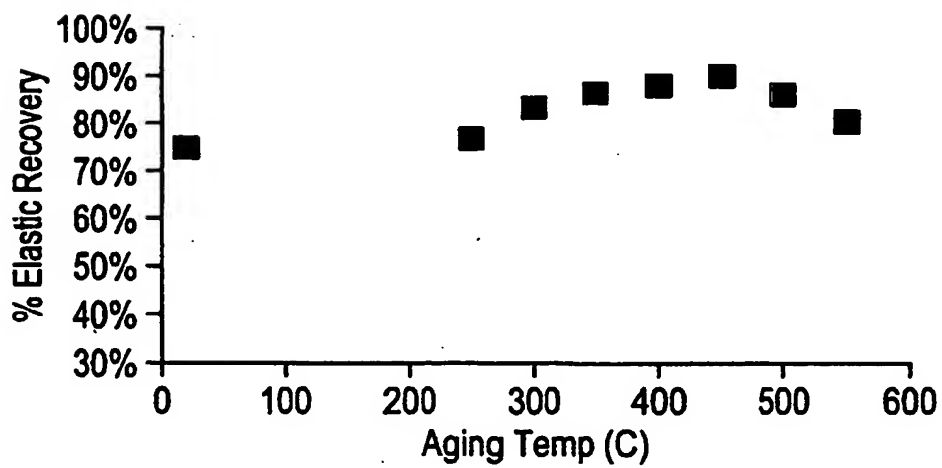


FIG. 6



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FIG. 7

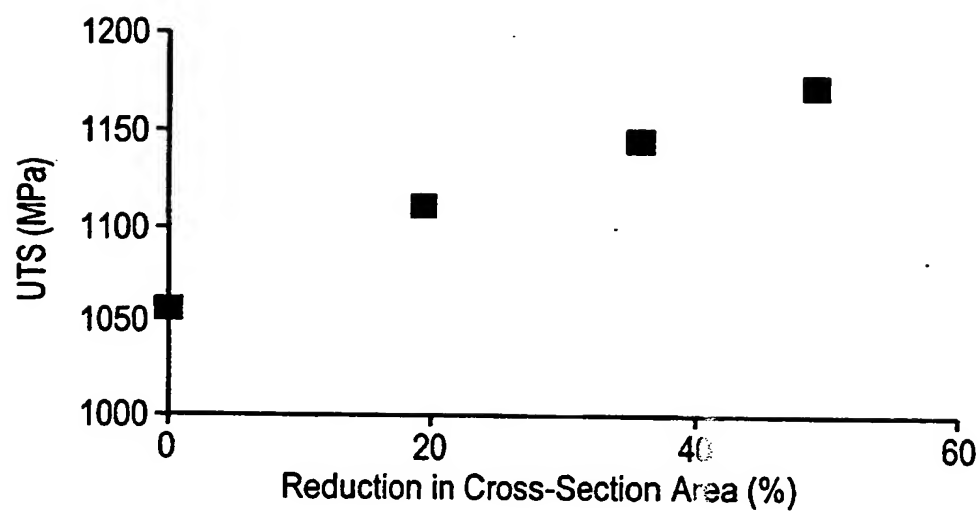
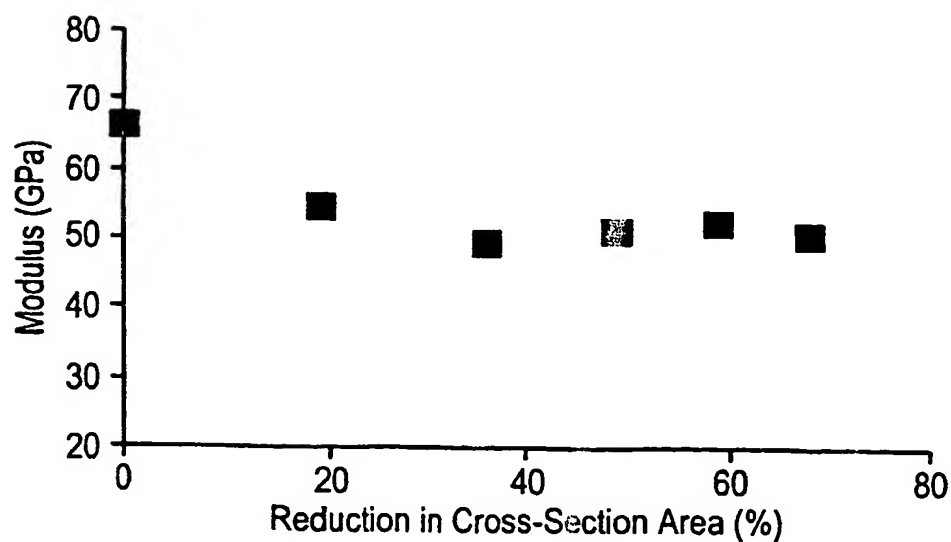


FIG. 8



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FIG. 9

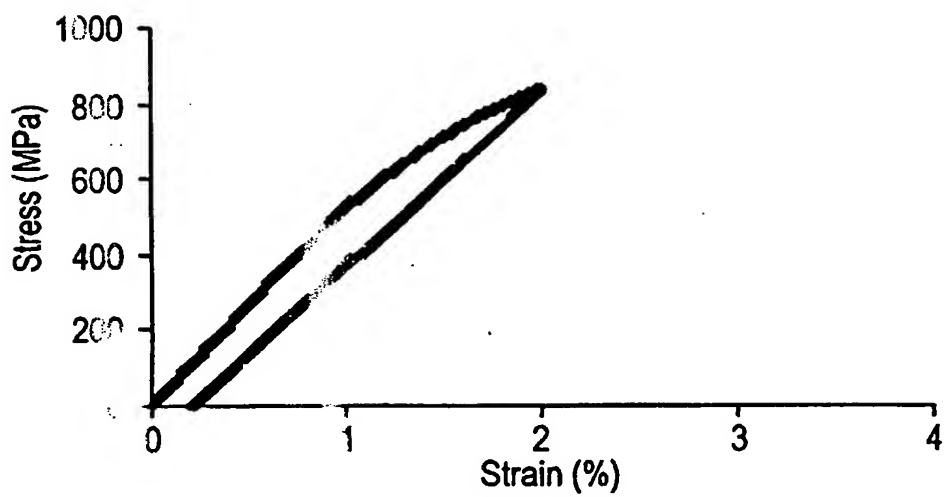
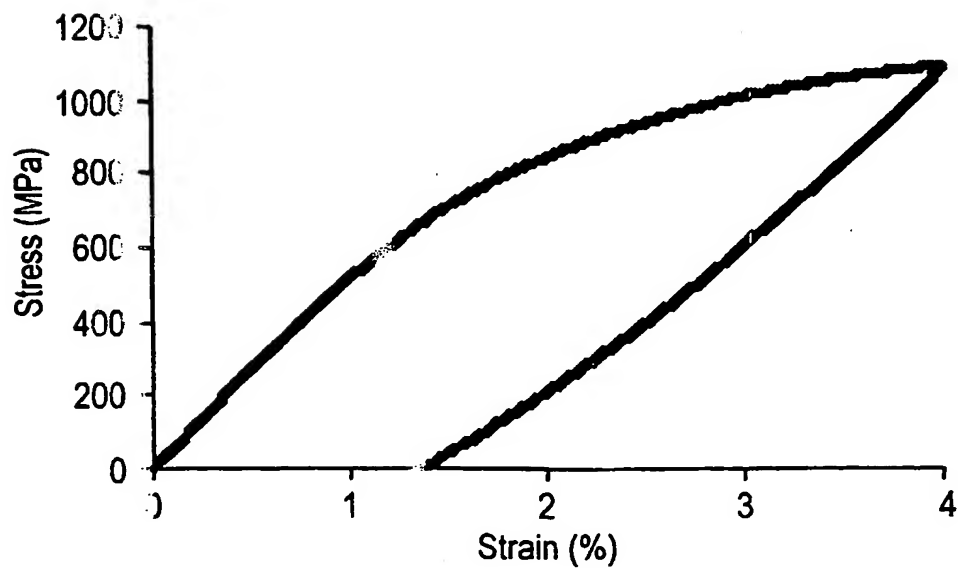


FIG. 10



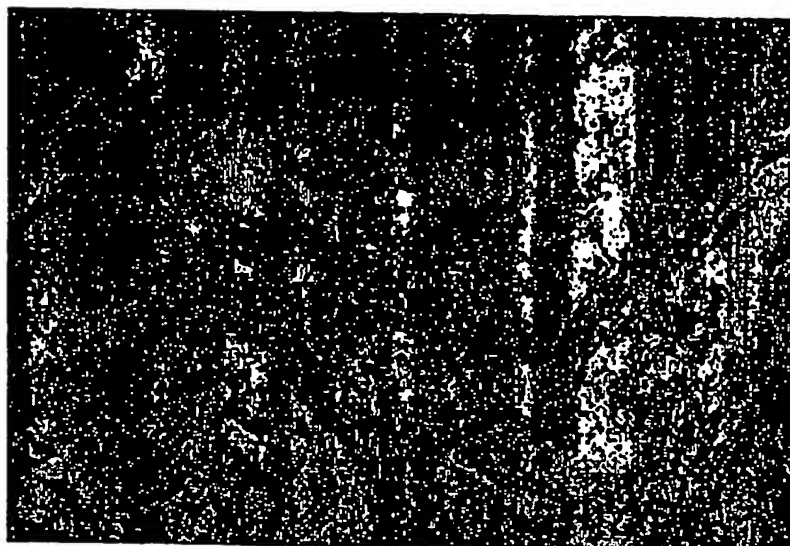


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FIG. 11



FIG. 12



SUBSTITUTE SHEET (RULE 26)

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FIG. 13

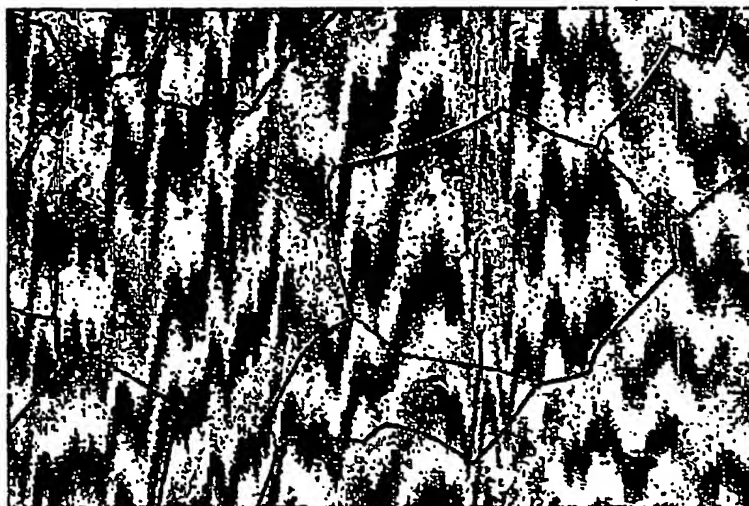
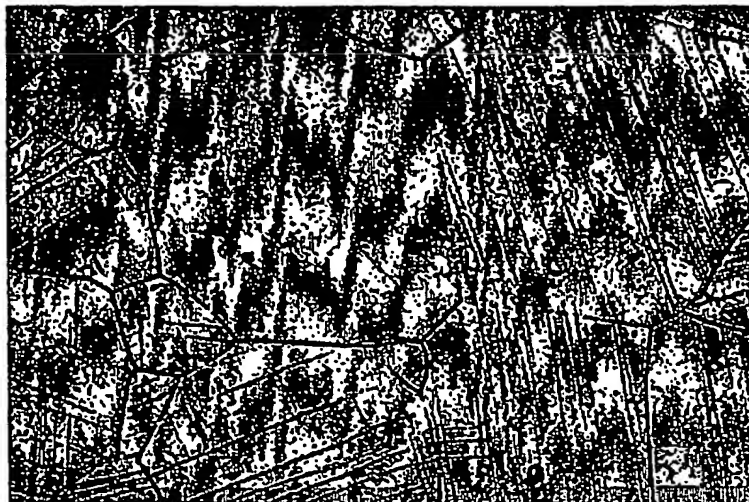


FIG. 14



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FIG. 15



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FIG. 16

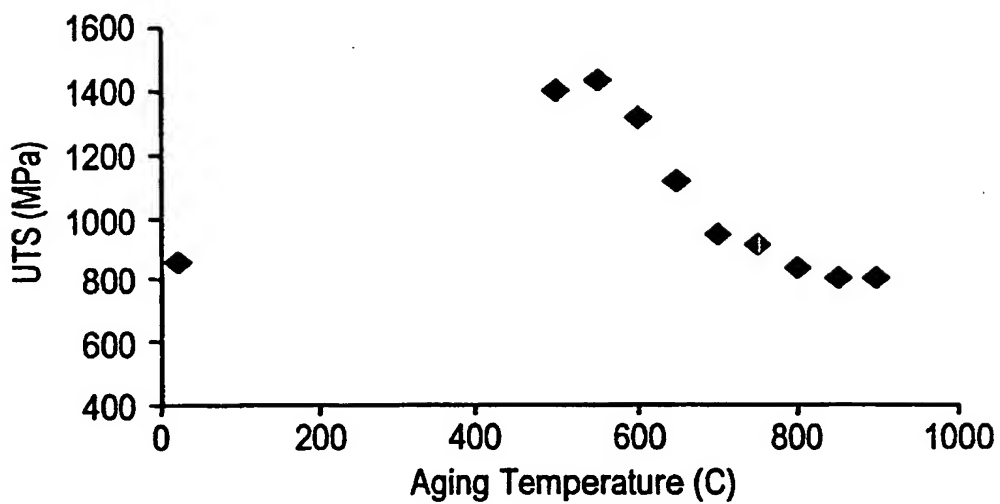
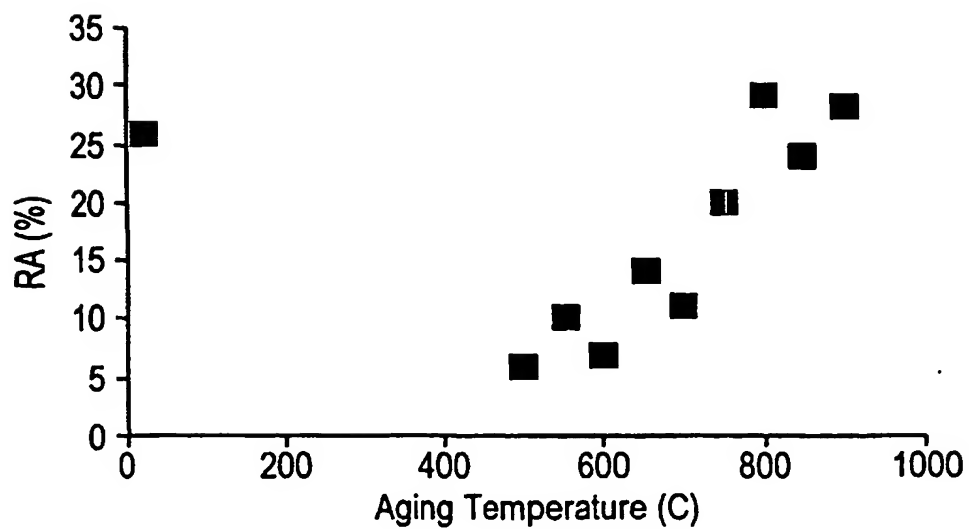


FIG. 17



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FIG. 18

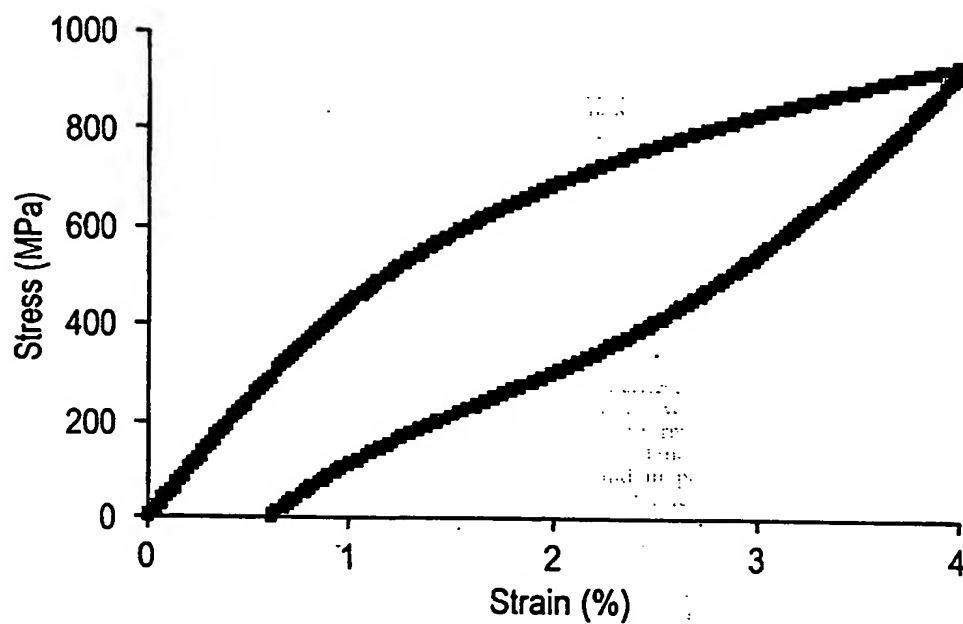
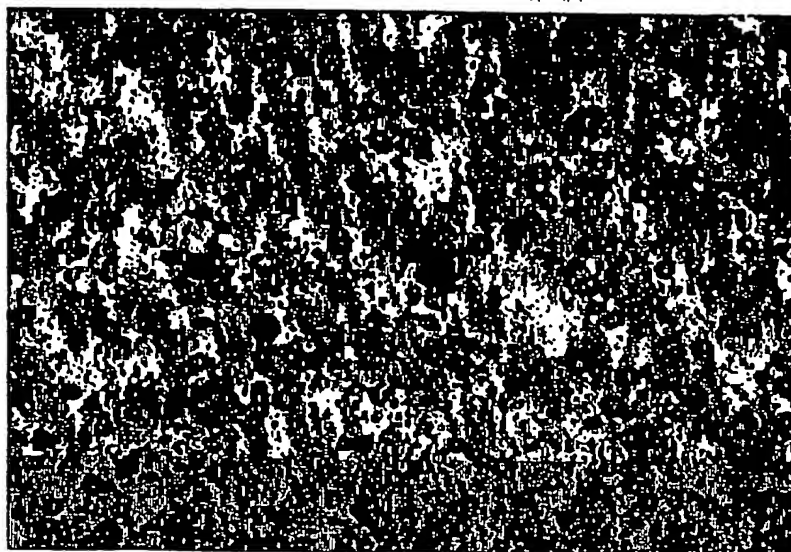


FIG. 19



# INTERNATIONAL SEARCH REPORT

International application No.

PCT/US03/20077

## A. CLASSIFICATION OF SUBJECT MATTER

IPC(7) : C22C 14/00; C22F 1/18

US CL : 148/421,671; 420/420

According to International Patent Classification (IPC) or to both national classification and IPC

## B. FIELDS SEARCHED

Minimum documentation searched (classification system followed by classification symbols)

U.S. : 148/421,671; 420/420

Documentation searched other than minimum documentation to the extent that such documents are included in the fields searched

Electronic data base consulted during the international search (name of data base and, where practicable, search terms used)  
EAST

## C. DOCUMENTS CONSIDERED TO BE RELEVANT

Category *	Citation of document, with indication, where appropriate, of the relevant passages	Relevant to claim No.
X	US 6,258,182 B1 (SCHETKY et al) 10 July 2001 (10.07.2001), Table III ex. 27, 28, 36,	1-8,13-23
---	column 7 line48, column 11 lines 15-18, column 12 lines 59-60.	-----
A		9-12



Further documents are listed in the continuation of Box C.



See patent family annex.

\* Special categories of cited documents:

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"E" earlier application or patent published on or after the international filing date

"L" document which may throw doubts on priority claim(s) or which is cited to establish the publication date of another citation or other special reason (as specified)

"O" document referring to an oral disclosure, use, exhibition or other means

"P" document published prior to the international filing date but later than the priority date claimed

"T"

later document published after the international filing date or priority date and not in conflict with the application but cited to understand the principle or theory underlying the invention

"X"

document of particular relevance; the claimed invention cannot be considered novel or cannot be considered to involve an inventive step when the document is taken alone

"Y"

document of particular relevance; the claimed invention cannot be considered to involve an inventive step when the document is combined with one or more other such documents, such combination being obvious to a person skilled in the art

"&"

document member of the same patent family

Date of the actual completion of the international search

04 November 2003 (04.11.2003)

Date of mailing of the international search report

28 NOV 2003

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